Frequency-based detection of replay attacks: application to a quadrotor UAV

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***Abstract*— Unmanned aerial vehicles (UAVs) are reported to be highly exposed as possible targets of cyber attacks, due to their strong strategic and economic value, and their increasing use in a wide range of operations. Among the most critical cyber attacks, replay attacks are performed by replacing the real data coming from the sensors with previously recorded data. In this way, the attacker may provoke performance degradation, cause instability, and allow to perform undetectable physical attacks. The main contribution of this paper is to investigate the applicability of a frequency-based detection method, which uses a sine wave with time-varying frequency as authentication signal, to a UAV affected by replay attacks. The effectiveness of the method is illustrated through simulation scenarios.**

I. Introduction

In recent years, the increasing integration of cyber (computation, communication, etc.) and physical processes has led to the introduction of a new class of systems, known as *cyber-physical systems* (CPSs) [1]. CPSs are characterized by a higher efficiency, but also by bigger vulnerabilities, which can be exploited by a malicious agent in order to perform a cyber attacks, resulting in some kind of damage or economical loss [2], [3].

Unmanned aerial vehicles (UAVs) are reported to be highly exposed as possible targets of cyber attacks, due to their strong strategic and economic value, and their increasing use in a wide range of operations, such as border surveillance, reconnaissance, transport, and even civilian tasks, as e.g. fun by hobby enthusiasts [4]. Among the attacks reported for UAVs, we recall the one performed in 2012 during a flight test in South Korea, in which a GPS jamming attack was executed on a S-100 Camcopter, resulting in a crash into the ground control van, which lead to the killing of an engineer and the injuring of two remote pilots [5].

Due to the importance of assessing the vulnerabilities of UAVs, and developing new defense mechanisms, which could help in mitigating them, simulations of cyber attacks on UAVs are being executed [6]. [7] performed a GPS spoofing attack on a quadrotor, resulting in the UAV landing at an incorrect location. A deauthentication attack on a quadrotor was used by [8] to demonstrate its vulnerability, resulting in the loss of control by the operator. The work [9] showed that a hacker can exploit shared data and predictable collision avoidance properties in order to control and alter the UAV's trajectory.

Replay attacks are critical cyber attacks carried out in two phases. In the first phase, the attacker records the real values provided by the sensors When an attack of this type is carried out, at first the attacker records the measurements coming from the sensors. Later, the attacker uses the recorded data to substitute the real data, which can lead to performance degradation, instability, and might allow to perform undetectable physical attacks. The threats to the security of UAVs brought by this type of attack have been well documented, see e.g. [10], [11], and several approaches have been proposed for detection purposes, which may be classified roughly into two categories: i) watermarking-based approaches [12]–[16], in which an authentication signal is introduced into the control signal, and the sensor outputs are analyzed to check if there is the effect of the introduced signal on the physical system; and ii) alternative methods [17], [18], which try to detect replay attacks without injecting signals in the control input.

The work [16] proposed to apply a frequency-based signature to detect replay attacks, and its application to a multiple tank system. The main contribution of the present paper is to investigate its applicability to a UAV affected by replay attacks. In particular, a sine wave with time-varying frequency is introduced into the closed-loop system to act as authentication signal and, by comparing the energies of appropriate signals, it is checked whether the received outputs are compatible with the authentication signal or not.

The remaining of the paper has the following structure. In Section II, the frequency-based replay attack detection technique is presented. In Section III, the proposed method is applied to a quadrotor. Section IV presents the simulation results. Finally, Section V provides the conclusions.

II. Replay attack detection method

A. Attack definition and overview of the detection method

Replay attacks corrupt the measurements coming from the sensors, and are carried out in two stages [12]:

1) in the first stage, the attacker records sensor data $y(t)$ in a time window $[t_0, t_0 + w]$, where w denotes the attack duration, without compromising their integrity;
2) in the second stage, the recorded data are used to replace the real data, such that the controlled system is exposed to some kind of harm without the supervisory unit/agent

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being aware of it, e.g., the system could be driven to a different operating point, or some act of theft/sabotage could be performed.

Hereafter, we recall the method proposed in [16], which detects replay attacks (also identifying which output channels are being attacked) by introducing a sine wave with time-varying frequency into the system. Then, the detector makes a decision by checking if the received outputs are compatible with the signature or not. More specifically, by applying a dynamical decoupling technique based on *vector fitting* [19], it can be ensured that a given input channel will affect only an output channel, which allows identifying which channel is being attacked. Finally, the detector compares the energies of band-pass signals to infer about the presence of a replay attack.

B. Signal generation

Consider a continuous-time linear time invariant system:

$$\dot{x}(t) = Ax(t) + Bu(t) \tag{1}$$

$$y(t) = Cx(t) \tag{2}$$

where $x \in \mathbb{R}^n$ is the state, $u \in \mathbb{R}^m$ is the input, $y \in \mathbb{R}^p$ is the output, and A, B, C denote matrices of appropriate dimensions. The system (1)-(2) is controlled by means of a state-feedback control law:

$$u(t) = -Kx(t) \tag{3}$$

such that the closed-loop system is described by:

$$\dot{x}(t) = (A - BK)x(t) \tag{4}$$

The frequency-based method for replay attack detection introduces a signature $\varsigma(t)$ into the input $u(t)$

$$u(t) = -Kx(t) + \varsigma(t) \tag{5}$$

It is straightforward to see that $\varsigma(t)$ will affect the output $y(t)$ according to the transfer function:

$$G(s) = C\left(sI - A + BK\right)^{-1} B \tag{6}$$

which usually contains coupling, i.e. a signature signal $\varsigma_i(t)$ introduced in the i-th input channel will contribute to the response of all the output signals contained in $y(t)$. Since we would like to use the signature signals to identify which channel is affected by the replay attack, we introduce a decoupler $F(s)$ with input $\tilde{\varsigma}(t)$ and output $\varsigma(t)$, such that the series interconnection $G_d(s) = G(s)F(s)$ is a decoupled system, i.e. made up by a diagonal transfer matrix. Then, we can introduce frequency-varying sinusoidal signals:

$$\tilde{\varsigma}_i = \tilde{\alpha}_i \cos\left(\omega_{\sigma_i(t)} t\right) \quad i = 1, \ldots, m \tag{7}$$

into the decoupler, where $\tilde{\alpha}_i$ denotes the signals' magnitudes, and $\sigma_i(t)$ denote piecewise constant signals, which take values between 1 and N, so that at each time $\omega_{\sigma_i(t)}$ take values within a set of predetermined frequencies $\{\omega_1, \ldots, \omega_N\}$, according to the value of $\varsigma_i(t)$. The idea behind introducing (7) is that by analyzing the output signals, we can decide about the attack occurrence by checking if the received outputs are compatible with the introduced signals.

It is clear that, since a complete decoupling of $G(s)F(s)$ is very demanding, and in some cases cannot be achieved at all [20], we can require decoupling for the finite set of frequencies ω_i, $i = 1, \ldots, N$. More specifically, by requiring that $F(\imath\omega_i) = G(\imath\omega_i)^{-1}$, we obtain N conditions that the decoupler should satisfy. Then, the robust numerical method for rational approximation known as *vector fitting* can be applied [21].

C. Detection logic

The output signal $y(t)$ will contain the effects of the different inputs to the process, e.g. the input signal $u(t)$. With the aim of analyzing only the content due to the signature signal $\tilde{\varsigma}(t)$, $y(t)$ is processed through a bank of band-pass filters (each filter corresponding to a different frequency ω_i, $i = 1, \ldots, N$), whose transfer functions are obtained as follows [22]:

$$H_i(s) = diag\left\{ \frac{\frac{\omega_i}{Q_i} s}{s^2 + \frac{\omega_i}{Q_i} s + \omega_i^2} \right\} \tag{8}$$

where Q_i is the selectivity of the filter, bigger values of which correspond to a narrower frequency response, but also a slower time response.

Denote as $z_{il}(t)$ the output of the filter $H_i(s)$ fed by $y_l(t)$. Then, we can obtain an estimate $\hat{\sigma}_l(t)$ of $\sigma_l(t)$ by comparing the different signals $z_{il}(t)$. As proposed in [16], a possible way to obtain $\hat{\sigma}_l(t)$ is by comparing the energies of $z_{il}(t)$ over the largest period associated to ω_i:

$$T_\omega = \max_{i=1,\ldots,N} \frac{2\pi}{\omega_i} \tag{9}$$

while also taking into account the time t_{trans} needed after a switch in $\sigma_l(t)$ in order for the transient to become neglectable. Then, the signals $\hat{\sigma}_l(t)$ are calculated as (10) (see top of the next page), where $t_s^* = \lfloor t/T_s \rfloor T_s$ denotes the last switching time. Then, the algorithm will compare the signals $\sigma_l(t)$, which are known, with the estimated signals $\hat{\sigma}_l(t)$. If a mismatch is found, a warning about $y_l(t)$ being affected by a replay attack is provided by the algorithm.

III. Application to a quadrotor

In this section, we present the application of the frequency-based replay attack detection method to a quadrotor.

A. Linearized model

The quadrotor is a vehicle that has four rotors in a cross configuration. Two rotors can rotate in a clockwise direction, while the other two can rotate anticlockwisely. The four rotors are controlled independently, and their speeds affect the overall movement of the quadrotor. It is assumed that the quadrotor has a rigid and symmetrical structure, with center of gravity that coincides with the body fixed frame, and rigid propellers, which provide thrust and drag forces that are proportional to the square of the propellers' speeds.

Let us introduce the state vector $x = \left[z, \dot{z}, \phi, \dot{\phi}, \theta, \dot{\theta}, \psi, \dot{\psi}\right]^T$ and the input vector $u = \left[\Omega_1, \Omega_2, \Omega_3, \Omega_4\right]^T$, where z is the

$$\hat{\sigma}_l(t) = \begin{cases} \sigma_l(t) & if\ \sigma_l(t) \neq \sigma_l(t-T_s)\ \wedge\ t \in [t_s^*, t_s^* + t_{trans} + T_\omega] \\ \arg\max_{i=1,\ldots,N} \int_{t-T_\omega}^{t} |z_{il}(\tau)|^2 d\tau & \text{otherwise} \end{cases} \tag{10}$$

height, ϕ, θ, ψ are the roll, pitch and yaw angle, respectively, and $\Omega_1, \Omega_2, \Omega_3, \Omega_4$ are the rotor speeds. By applying a constant feedforward input $u_e = [\Omega_{1,e}, \Omega_{2,e}, \Omega_{3,e}, \Omega_{4,e}]^T$, with:

$$\Omega_{1,e} = \Omega_{2,e} = \Omega_{3,e} = \Omega_{4,e} = \sqrt{\frac{gm}{4b}} \tag{11}$$

a hovering equilibrium point is obtained, characterized by a constant equilibrium state x_e (in particular, the equilibrium height is a value z_e, and the equilibrium angles ϕ_e, θ_e and ψ_e are all zero). Then, by considering deviations Δx of the state variables from x_e, and performing a linearization of the nonlinear state equations (see [23], [24]), a model akin to (1) is obtained:

$$\Delta\dot{x}(t) = A\Delta x(t) + B\Delta u(t) \tag{12}$$

with:

$$A = \begin{bmatrix} 0 & 1 & 0 & 0 & 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 1 & 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 & 0 & 1 & 0 & 0 \\ 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 & 0 & 0 & 0 & 1 \\ 0 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \end{bmatrix}$$

$$B = \begin{bmatrix} 0 & 0 & 0 & 0 \\ \sqrt{g\frac{b}{m}} & \sqrt{g\frac{b}{m}} & \sqrt{g\frac{b}{m}} & \sqrt{g\frac{b}{m}} \\ 0 & 0 & 0 & 0 \\ 0 & -\frac{l}{I_x}\sqrt{gmb} & 0 & \frac{l}{I_x}\sqrt{gmb} \\ 0 & 0 & 0 & 0 \\ -\frac{l}{I_y}\sqrt{gmb} & 0 & \frac{l}{I_y}\sqrt{gmb} & 0 \\ 0 & 0 & 0 & 0 \\ -\frac{d}{I_z}\sqrt{g\frac{m}{b}} & \frac{d}{I_z}\sqrt{g\frac{m}{b}} & -\frac{d}{I_z}\sqrt{g\frac{m}{b}} & \frac{d}{I_z}\sqrt{g\frac{m}{b}} \end{bmatrix}$$

which, together with a linear error-feedback control law $\Delta u(t) = -K\Delta x(t)$, leads to:

$$\Delta\dot{x}(t) = (A - BK)\Delta x(t)$$

We will assume that both the altitude and the attitude of the quadrotor are monitored by a ground station. The communication between the UAV and the station can be hacked through a replay attack. Hence, the system's description (1)-(2) is completed by the output matrix C, which is given by:

$$C = \begin{bmatrix} 1 & 0 & 0 & 0 & 0 & 0 & 0 & 0 \\ 0 & 0 & 1 & 0 & 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 & 1 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 & 0 & 0 & 1 & 0 \end{bmatrix}$$

or, equivalently, by defining $\Delta y(t) = y(t) - [z_e, \phi_e, \theta_e, \psi_e]^T$:

$$\Delta y(t) = C\Delta x(t)$$

TABLE I
SYSTEM PARAMETERS VALUES

Parameter	Value	Unit	Parameter	Value	Unit
I_x	$8.1e-3$	Nms^2	J	$104e-6$	Nms^2
I_y	$8.1e-3$	Nms^2	l	0.24	m
I_z	$14.2e-3$	Nms^2	b	$54.2e-6$	Ns^2
m	1	kg	d	$1.1e-6$	Nms^2
g	9.81	ms^{-2}			

Also, for the remaining of the manuscript, we will assume that the quadrotor's parameters are the ones provided in Table I, where I_x, I_y, I_z denote the body inertia along x, y, z axes, m is the quadrotor's mass, g denotes the gravity acceleration, J is the propeller inertia, l is the lever's length, and b and d are the thrust and drag factor, respectively.

B. Choice of the design parameters

Consider a detector, as described in the previous section, with $N = 2$ and $\omega_2 = 2\omega_1$. Due to the fact that by design:

$$F(\imath\omega_i) \approxeq G(\imath\omega_i)^{-1} = \left[C(\imath\omega_i - A + BK)^{-1}B\right]^{-1} \tag{13}$$

the following relationship holds between the magnitudes $\tilde{\alpha}_1$, $\tilde{\alpha}_2$, $\tilde{\alpha}_3$, $\tilde{\alpha}_4$ of $\tilde{\varsigma}_1$, $\tilde{\varsigma}_2$, $\tilde{\varsigma}_3$, $\tilde{\varsigma}_4$ (input signals to the decoupler) and the maximum magnitudes $\bar{\alpha}_1$, $\bar{\alpha}_2$, $\bar{\alpha}_3$, $\bar{\alpha}_4$ of ς_1, ς_2, ς_3, ς_4 (outputs of the decoupler, which are inputs of the actuators);

$$\begin{bmatrix} \bar{\alpha}_1 \\ \bar{\alpha}_2 \\ \bar{\alpha}_3 \\ \bar{\alpha}_4 \end{bmatrix} = \max_{i=1,2} \left| \left[C(\imath\omega_i - A + BK)^{-1}B\right]^{-1} \right| \begin{bmatrix} \tilde{\alpha}_1 \\ \tilde{\alpha}_2 \\ \tilde{\alpha}_3 \\ \tilde{\alpha}_4 \end{bmatrix} \tag{14}$$

where the max symbol is meant element-wise. By requiring that:

$$\bar{\alpha}_l \leq \kappa_l u_{e,l}, l = 1, \ldots, n_u \tag{15}$$

where $\kappa_l \ll 1$ and $u_{e,l}$ is the l-th element of u_e, such that the increment of the input signal brought by the signature signal is relatively small, a set of feasible frequencies is calculated, and ω_1 is chosen as the maximum value among these frequencies. The values $\tilde{\alpha}_l$, $l = 1, \ldots, 4$ are design parameters whose values should be chosen small enough such that $\tilde{\varsigma}(t)$ is not clearly visible in the output signals, and big enough to overcome the effect of exogenous disturbances and measurement noise. In the following, we select $\kappa_1 = \kappa_2 = \kappa_3 = \kappa_4 = 1/10$ and $\tilde{\alpha}_1 = 0.005$ and $\tilde{\alpha}_2 = \tilde{\alpha}_3 = \tilde{\alpha}_4 = 0.01$.

Four different controllers K_1, K_2, K_3, K_4 have been considered in order to study how the particular choice of the controller affected both the set of feasible frequencies for ω_1 and the decoupling performance of the decoupler $F(s)$. All the controllers have been designed by pole placement, by requiring that the closed-loop poles were placed in different positions of the complex plane, as resumed in Table II.

TABLE II

CLOSED-LOOP POLES POSITION

Controller	Dominant poles	Fast poles
K_1	$\{-3,-3,-3,-3\}$	$\{-10,-10,-10,-10\}$
K_2	$\{-3,-3,-3,-3\}$	$\{-20,-20,-20,-20\}$
K_3	$\{-3,-3,-3,-3\}$	$\{-50,-50,-50,-50\}$
K_4	$\{-3,-3,-3,-3\}$	$\{-100,-100,-100,-100\}$

Eq. (14) has been used to calculate the expected $\bar{\alpha}$ (due to symmetry of the quadrotor, $\alpha_1 = \alpha_2 = \alpha_3 = \alpha_4$) for the different controllers, such that (15) can be checked graphically, as shown in Fig. 1. Using the controller K_1, a value $\omega_1 = 5.1 rad/s$ has been found, while with the controllers K_2 and K_3, $\omega_1 = 3.3 rad/s$ and $\omega_1 = 0.3 rad/s$ have been obtained. On the other hand, no feasible frequency ω_1 was found for controller gain K_4. Although in principle it is desired to obtain a value of ω_1 as high as possible, so that T_ω in (9) becomes smaller, and the detection algorithm based on estimating $\hat{\sigma}_l(t)$ using (10) behaves faster, the choice of ω_1 should take into account the decoupler performance which, looking at (13), depends on the specific controller gain (the decoupler can be obtained applying the VFIT3 routine[1] [19], [25], as described in [16]).

In particular, Fig. 2 shows the Bode plot of the transfer function from the input 1 to the output 1 of the cascade $G(s)F(s)$, for the different controller gains K_1, K_2, K_3. Note that the values on the x-axis are normalized with respect to ω_1, such that $\omega_{norm} = 1 rad/s$ corresponds to $\omega = 5.1 rad/s$, $\omega = 3.3 rad/s$ and $\omega = 0.3 rad/s$, respectively. It can be seen that the best decoupling performance is achieved with the controller K_3. However, using K_3 would lead to $\omega_1 = 0.3 rad/s$ and, through (9), to $T_\omega = 20.94 s$, ultimately leading to a very slow attack detector. For this reason, using K_2 ($\omega_1 = 3.3 rad/s$, $T_\omega = 1.91 s$) is deemed to be a good tradeoff between a fast detector and a good decoupling, and will be the controller used in the next section. By requiring an attenuation of $-20 dB$ at frequency ω_2 for the first band-pass filter, and at frequency ω_1 for the second band-pass filter, the following value is obtained for the selectivity parameters: $Q_1 = Q_2 = 2\sqrt{11}$. Also, t_{trans} is chosen as the biggest among the settling times of the band-pass filters $H_i(s)$, and T_s is chosen as $T_s = 4t_{trans}$, as in [16], thus obtaining $t_{trans} = 15.87 s$ and $T_s = 63.48 s$. For the sake of completeness, Fig. 3 compares the Bode plot of the non-decoupled system (blue line) and the Bode plot of the decoupled system (red line). The series interconnection $G(s)F(s)$ approximates an identity matrix, so that a good decoupling has been achieved.

IV. SIMULATION RESULTS

To demonstrate the effectiveness of the proposed strategy, two different scenarios are considered. In Scenario 1, the first output measurements are recorded by the attacker during the first $200 s$, and then replayed periodically starting from $t = 200 s$. The arising mismatch between the outputs of the band-pass filters $z_{11}(t)$, $z_{21}(t)$ and the time-varying frequency

[1]https://www.sintef.no/projectweb/vectfit

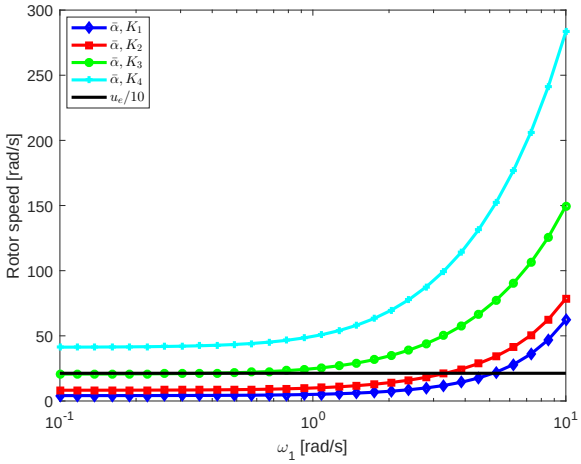

Fig. 1. Graphical check of condition (15).

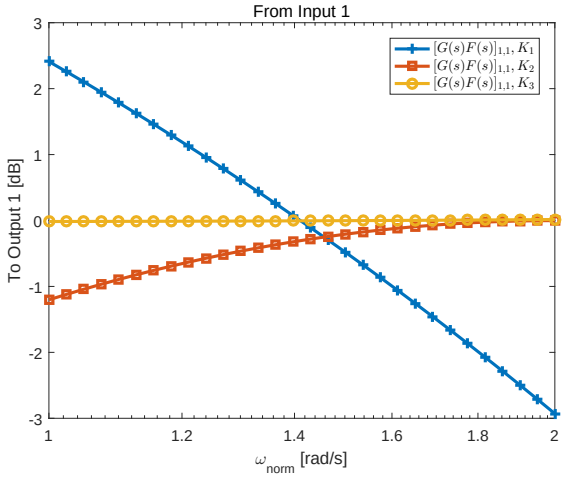

Fig. 2. Decoupling performance (Bode plot of the transfer function from input 1 to output 1 for the cascade $G(s)F(s)$, using different controller gains).

profile of $\omega_{\sigma_1(t)}$ (see Fig. 4) allows detecting the attack after $8.21 s$, as shown in Fig. 5, which depicts the logical assessment of $\sigma_i(t) == \hat{\sigma}_i(t)$, $i = 1,2,3,4$. In Scenario 2, the attacker corrupts the measurements $y_2(t)$ and $y_3(t)$ instead. In this case, the mismatches $\hat{\sigma}_2(t) \neq \sigma_2(t)$ and $\hat{\sigma}_3(t) \neq \sigma_3(t)$ allow identifying correctly the output channels as being attacked at time $398.65 s$ and $208.21 s$, respectively (see Figs. 6-7).

V. CONCLUSIONS

This work has studied the applicability of a frequency-based replay attack detection method to a quadrotor UAV. In particular, a sine wave with time-varying frequency is introduced into the closed-loop system and, by comparing the energies of appropriate signals, it is checked if the time profile of the frequency components in the output signals are compatible with the authentication signal. The simulation scenarios have shown that the method can be applied to the

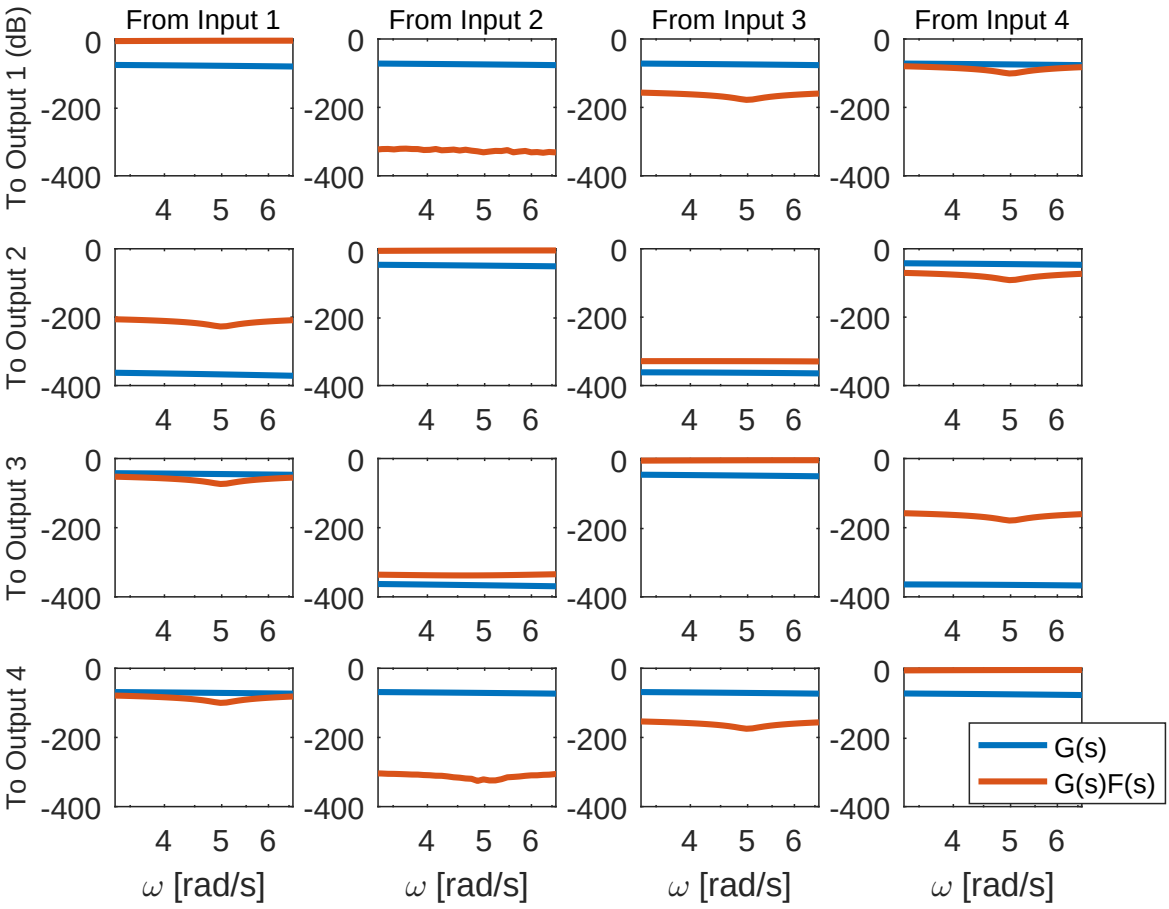

Fig. 3. Decoupling performance (Bode plots).

considerate case study, being capable of not triggering false alarms and identifying correctly the attacked output channels.

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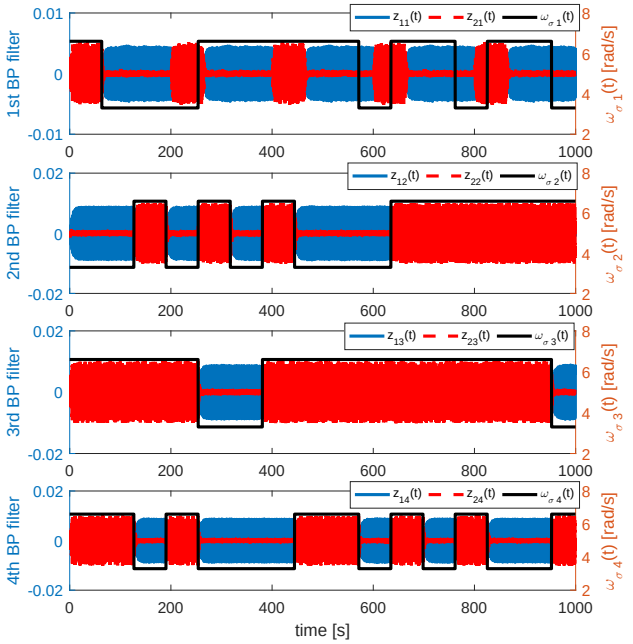

Fig. 4. Scenario 1. Outputs of $z_{il}(t)$ and time-varying frequency $\omega_\sigma(t)$.

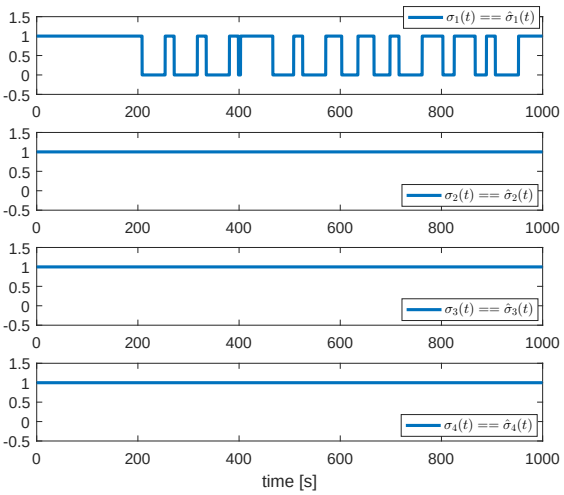

Fig. 5. Scenario 1. Replay attack detection test.

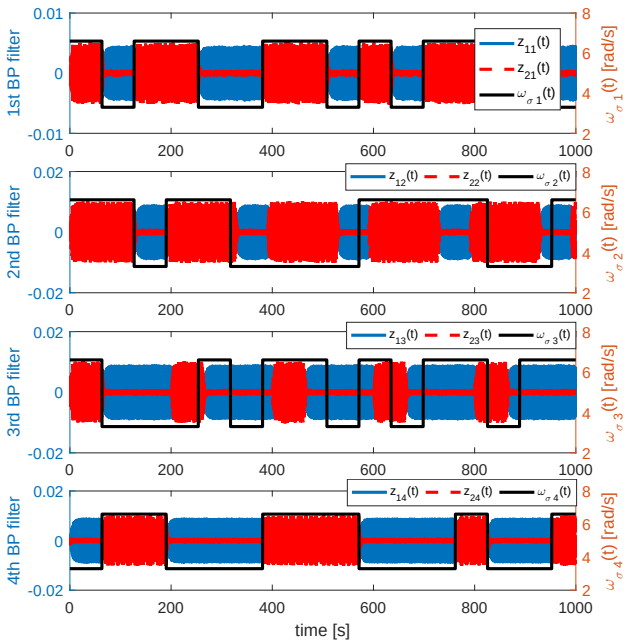

Fig. 6. Scenario 2. Outputs of $z_{il}(t)$ and time-varying frequency $\omega_\sigma(t)$.

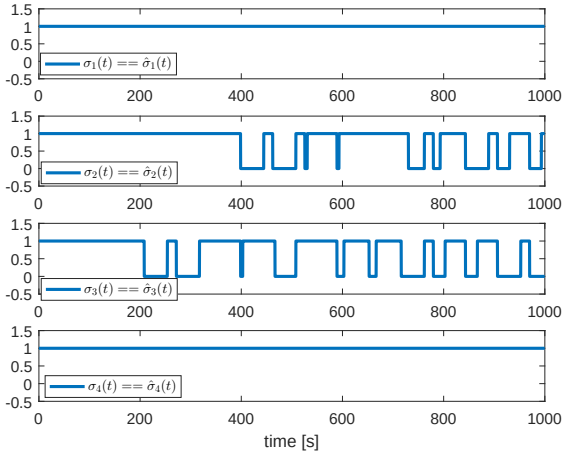

Fig. 7. Scenario 2. Replay attack detection test.

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